

CFD ANALYSIS OF WIND TURBINE BLADE FOR LOW WIND SPEED

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Abstract - The power generation problem is one of the important aspects of the world that has to address with high level of scientific reasoning and profound knowledge in the field of energy sources. Power can also be obtained from renewable resources, which include hydro power plants, solar power plants, solar thermal power plants and most importantly wind turbines. Hence harnessing and exploiting the energy comprised in the wind is important. Development of horizontal axis wind turbines (HAWTs) for small-scale application is important. A six bladed wind turbine system using the airfoils- Gottingen 364,384,428,480 and 682 were taken for the research. Using the design knowledge of on airfoils, key factors lift, drag and stall of the airfoil where studied and twist of the blade segments are designed with precision. Hence, the blades for a six bladed wind turbine system is designed for the optimal coefficient of lift value using Q-blade software with careful consideration at each location. Further, the aerodynamic parameters i.e. drag force, lift force, coefficient of drag, coefficient of lift and lift to drag ratio., pertaining to the wind turbine, the necessary calculations are formulated with the aid of Computational Fluid dynamics (CFD). Using the appropriate procedure of K- ϵ method to solve for the wind flow over the blade in an enclosed medium, the output parameters are calculated from the flow pattern in two functional form. By this method it is possible to reach the maximum amount of accuracy in carrying out the analysis and one can also obtain a very detailed output results. To check the aerodynamic design performance, the six bladed wind turbine system is validated using Numerical Study Computational Fluid Dynamics (CFD) using standard software's (ANSYS-FLUENT) which forecast the performance of blades at low wind velocities. The parameters which are taken for analysis are carefully checked and correlated with IEC- 61400 Standard of small wind turbine system. The average wind velocity at hub height 3m/s is taken as wind designed input and the proposed designed six bladed wind turbine for each airfoils is carried for analysis. A six bladed wind turbine system having Gottingen airfoils 364,384,428,480 and 682 respectively were studied.

Key Words- Gottingen, MRF, Mesh, Contour, K- ϵ .

1. INTRODUCTION

Wind Energy is the exercising of air flow through wind turbines to automatic power generators for electric power is a handy source of renewable energy. Wind power has been harnessed since before the Industrial Revolution in the form of windmills to thresh grains and for water pumping applications. The gradual and unceasing growth of this technology has led to the expansion of power generating massive wind turbines that we see these days. Wind Turbines largely, are huge in dimensions, noisy and require a wide space to function. This calls for installation of Wind Turbines in uninhabited rural areas, which are categorized by a robust flow of wind. Micro wind turbines can help to trim down high electricity costs and provide a cushion against ever increasing electricity demands. At remote locations in India, a micro wind turbine would be a practical and economical solution for the electricity complications, which might reduce our dependence on fossil fuels and circumvent greenhouse gases. Depending on the local wind resource and utility rates, a small wind energy system can minimize a customer's electricity bill by 50% to 60%. It can be installed as a standalone system, eliminating the high cost of extending utility power lines to a remote location, or it can be connected to the power grid, enabling the customer to sell excess power to the grid or buy additional power when needed through a reverse metering system. Payback on capital investment is within the life span of the wind turbine. Given that a large portion of the Indian land will not be viable for the use of traditional windmills due to low wind speeds, efficient blades and a generator that would produce energy even at very low wind 2 speed are requisite. As the transmission losses in India are very high, placing the turbine near the place of power consumption could prove to be a feasible solution.

1.1 HISTORY OF AIRFOILS

That flat surfaces in the wind could produce the sideways force that we now call lift was a very ancient observation. Two early applications of it, the windmill and the fore-aft rigged sail, date back at least 800 years. It was also perfectly evident to any thinking person that what kept birds and bats aloft were the large flat surfaces attached to their arms. Neither the feathers of birds nor the fabric of sails and windmill blades had any thickness to speak of, and so the earliest lifting surfaces were just those surfaces. Thin surfaces restrained by a supporting structure naturally bellied out under air pressure, assuming what we now call a "cambered" -- that is, arched -- shape. The fact that camber was actually beneficial seems first to have been appreciated -- at least in writing -- by an English civil engineer of the 18th century, John Smeaton, who noted that curving the surfaces of their blades, improved the performance of windmills. For the next century and a half, nothing noteworthy occurred -- other than the invention of the modern airplane, in 1804, by another Englishman, George Cayley. Despite the random and ad

hoc quality of these early airfoil designs, efforts were being made to sort out the wheat from the chaff in wind tunnels. At first, unfortunately, investigators did not recognize the importance of scale. They tested very small models at very low speeds, and, because speed and size actually play important roles in the behavior of flowing air, their results supported the mistaken guess that thin airfoils were superior.

2. LITERATURE SURVEY

2.1. AERODYNAMICS OF WIND TURBINE

Wright & Wood et al (2004) observed and explained about starting a small horizontal axis wind turbine at low wind speed they studied behavior of blades could generate unexpected high torque. At the same time, the non-dimensional pitch rate and reduced frequency are too small to suggest a significant increase of the torque through the effects of unsteadiness. The torque subsequently decreased owing to inappropriate blade angles of attack, which lead to a substantial "idle time" at both high and low wind speeds, in which the rotating blades are accelerated only slowly and the angles of attack decreased slowly.

Sikandar Khan et al. (2012) had a work on the aerodynamic analysis and dynamic modelling of small horizontal axis wind turbine. This research work, mainly focus on dynamic analysis and Pitch control of horizontal axis wind turbine. Pitch control is one of the most useful and important control, present in all of the current wind turbine models. For dynamic analysis and pitch control, simulation techniques and experimental works were done side by side.

Habali (2000) et al gave an overview of the operation of wind turbines at low cut-in wind speeds and he able to find possible way through aerodynamic optimization of the rotor blades, which is the most important part of a wind turbine.

Chen (2011) et al made experimental study and he briefed about the Aerodynamic shape optimization, which is one of the main research fields that is directly related to power production of a wind turbine. The optimum distributions of the chord length and the pitch angle in each section can be acquired according to the design parameters, which include the rated wind speed, number of blades, design tip speed ratio and design angle of attack.

Arifujjaman et al (2008) modeled a small wind turbine with furling mechanism and investigated the model resulting in dynamics. He simulates by regulating the speed of the wind turbine via a load control method. Tip speed ratio and hill climbing control methods provides maximum power extraction from a small wind turbine. Small-scale wind 3 turbines installed within the built environment is classified as micro generation technology.

Ditkovich et al. (2014) had provided an overview on a method for estimating the fixed and variable speed turbine performance index. The approach is based on Weibull wind probability distribution function and manufacturer provided power curve.

Xu et al. (2013) gave a brief study on building and explained about miniature wind turbine which received great attention recently for powering low power devices with a physics based comprehensive model for predicting the performance of miniature horizontal axis wind turbine.

Dong Ok Yu & Oh Joon Kwon et al (2014) designed a small wind turbine generator system which can be mounted in coastal areas and islands slop with a good structures and a protective system for reliable operation. A small wind turbine prototype with high reliability is designed and examined in the real island conditions.

Johansen et al (2002) made a study on wind turbine blades and he investigated Flows near blade tip and hub numerically. He found the flow through an untwisted blade HAWT is quite complicated to solve numerically because of the rotation of the turbine, coupled with turbulence and stall effects, hence The rotating wind turbine can be modelled with static or dynamic grid method.

2.2. NUMERICAL STUDY ON BLADES

Douvi Eleni et al. (2012) made a study on the analysis of the two dimensional subsonic flow over a National Advisory Committee for Aeronautics (NACA) 0012 airfoil at various angles of attack and operating at a Reynolds number of 3×10^6 and he explained the behavior of the study. The flow was obtained by solving the steady-state governing equations of continuity and momentum conservation combined with one of three turbulence models (Spalart-Allmaras, Realizable and shear stress transport (SST)). This aims to the validation of the models through the comparison of the predictions and the free field experimental measurements for the selected airfoil.

McKittrick et al (2001) gave a detailed report on analyzing the CFD techniques and it is found CFD methodology's can effectively predict the load characteristics of wind turbines and it is reasonable to employ pressure distributions obtained from CFD calculations as load conditions to This work highlighted two areas in computational fluid dynamics (CFD) which require for investigation transition point prediction and turbulence modelling. The laminar to turbulent transition point was modelled in order to get accurate results for the drag coefficient at various Reynolds numbers.

Karna Patel et al. (2014) made a brief study on CFD analysis of a airfoil and he provided information on relationship of drag, lift and forces and it impact on airfoils. From the study, it is clear to find drag and lift forces using CFD methodology. The analysis of the two dimensional subsonic flow over a NACA 0012 airfoil at various angles of attack and operating at a Reynolds number of $3 \times E+06$ is made and the CFD simulation results show close agreement with those of the experiments, thus suggesting a reliable alternative to experimental method in determining drag and lift. He also explained about the wind tunnel testing method to determine airfoil lift and drag forces where the process is quite laborious & costly more than CFD techniques. Thus validation of the research work has gone through analytical method then validation by experimental testing.

Kevin Cox & Andreas Echtermeyer et al (2012) gave an overview on the structural design and analysis of a 4 10MW wind turbine blade. The structural aspects of a 70- meter-long blade in an upwind, horizontal-axis wind turbine were developed and build into use at high wind speed location. The structure of the blade was build using hybrid composites which have glass and carbon fiber plies for yielding a lightweight design with a low tip deflection. The blade was subjected to FEA studies to demonstrate its ability to withstand the extreme loading conditions as per the international offshore wind standard. The results confirmed the design to have acceptable performance with regard to tip deflection, maximum and minimum strains, and critical buckling load.

Wenlei Sun et al. (2010) gave an overview on the design and the research re-search difficulty in building a largescale wind turbine blades to work. Where the major drawbacks are in the work occupy in the blades designs because of threedimensional blades model.

Chalothorn Thumthae & Tawit Chitsomboon et al (2006) recommended the importance of optimal angle of attack for untwisted blade in wind turbine. Hence the numerical simulation of horizontal axis wind turbines (HAWTs) with untwisted blade was performed to determine the optimal angle of attack that produces the highest power output. The numerical solution was carried out by solving conservation equations in a rotating reference frame wherein the blades and grids were fixed in relation to the rotating frame. Computational results of the 12° pitch compare favorably with the field experimental data of The National Renewable Laboratory (USA), for both inviscid and turbulent conditions. Numerical experiments were conducted by varying the pitch angles and the wind speeds. The power outputs reach maximum at pitch angles: 4.12° , 5.28° , 6.66° and 8.76° for the wind speeds 7.2, 8.0, 9.0 and 10.5 m/s, respectively. The optimal angles of attack were then obtained from the data.

Amano & Ryan Malloy et al (2013) explained about the improvement of blade aerodynamics and explore the possibility of increasing the number of profitable sites by optimizing wind turbine blade design for low wind speed areas.

Pedro Bañuelos-Sánchez et al. (2011) he gave a overview on low power-low cost horizontal axis wind turbine for 350Watt application. He describes a design and implementation of a low power Horizontal Axis Wind Turbine where the axis of rotation is parallel to the ground. Rotor, blades, supporting hub, and drive train are designed using computational software. Implemented wind turbine was tested and experimental results were obtained for 3.5 m/s to 9 m/s wind speed.

2.3. SCOPE

- The scope of the research work is to design a six
- bladed small horizontal axis wind turbine system - suitable for
- low speed wind condition, which would be an alternative
- The source for conventional large diameter wind turbine blades.

2.4. OBJECTIVES

The main objectives of this research is:

- To design a wind turbine which is cost effective and generates high power output.
- To design the blade, so as to have an extraordinary self-starting capacity at very low wind speeds.

- To model the blade of the turbine using different airfoils.
- To perform a steady state analysis using ANSYS 14.5 for the various blades modeled.
- To perform an analysis on different airfoil blades and decide on the most efficient one for the final blades.
- Fabricate the wind turbine and yoke the theoretical values with the test results.

3. AERODYNAMIC DESIGN OF WIND TURBINE

Rotor design, airfoil selection and Blade design constitute the aerodynamic design of the blade. Dynamic behavior, strength and fatigue properties of the rotor have to be taken into consideration when a wind turbine is being designed and constructed. The Rotor tip is designed with great attention and precision as the tip of the blade moves substantially faster than the root of the blade. To select the suitable wind turbine, three design speed parameters, namely cut-in wind speed $V_{cut\ in}$, rated wind speed V_{rated} and cut out wind speed $V_{cut\ out}$ can be calculated by using equations.

$$v_{cut\ in} = 0.7\bar{u}$$

$$1.5\bar{u} \leq v_{rated} \leq 2.0\bar{u}$$

$$v_{cut\ out} \geq 3\bar{u}$$

Based on the average wind speeds in Tamil Nadu the rotor blades are to be designed. In this research, the annual average wind speed (\bar{u}) of Tamil Nadu is anticipated to be 3m/s. The cut in speed is 1.4m/s and the cut out speed is 9m/s for the average wind speed taken. This average wind speed is the rated speed of the wind turbine.

4. AIRFOIL SELECTION

The main purpose of choosing this airfoil when compared to other airfoils is to generate maximum lift force by minimizing drag force. This airfoil produced efficient results for wind turbines for different domestic applications. Lift force for the airfoil is high which is highly suited for low wind speed turbines. A higher L/D ratio is also possible with non- symmetrical profiles. This airfoil is chosen because it satisfies many of the design criteria for optimal wind turbine rotor performance.

5. AIRFOIL DESIGN METHODOLOGY

The process of airfoil design proceeds from a knowledge of the boundary layer properties and the relation between geometry and pressure distribution. The goal of an airfoil design varies. Some airfoils are designed to produce low drag (and may not be required to generate lift at all.) Some sections may need to produce low drag while producing a given amount of lift. In some cases, the drag doesn't really matter - it is maximum lift that is important. The section may be required to achieve this performance with a constraint on thickness, or pitching moment, or off-design performance, or other unusual constraints. The relative thickness of aero foil changes in the span direction. For wing, shape Max. Relative thickness, it decreases from 40% of root to 10%of leaf apex, to gain Max. Wind energy utilizing coefficient. The methods for airfoil design can be classified into two categories: direct and inverse design.

5.1. DIRECT DESIGN METHOD

The direct airfoil design method involves with specification of respective airfoils which are geometry sections, Calculated pressure and performance. Using the above parameters or specifications of the airfoil, its later tends to evaluates the given shape and then modifies the shape to improve the performance. The two main sub problems in Direct method of airfoil analysis are

- The identification of the measure of Performance.

- The approach to changing the shape so that the performance is improved

The simplest form of direct airfoil design involves starting with an assumed airfoil shape (such as a NACA airfoil), determining the characteristic of this section that is most problematic, and fixing this problem. The process involves fixing the most obvious problems with a given airfoil and it is repeated until no major problem with the section is able to be found. The design of such airfoils does not require a specific definition of a scalar objective function, but it requires some expertise to identify the potential problems and often considerable expertise to fix them. Due to the complication and expertise involvement, the present research work for airfoil design uses inverse design of airfoils which is discussed below.

5.2. INVERSE DESIGN METHOD

Another type of objective function is the target pressure distribution. It is sometimes possible to specify a desired C_p distribution and use the least squares difference between the actual and target C_p 's as the objective. This is the basic idea behind a variety of methods for inverse design. As an example, thin airfoil theory can be used to solve for the shape of the camber line that produces a specified pressure difference on an airfoil in potential flow. The second part of the design problem starts when one has somehow defined an objective for the airfoil design. This stage of the design involves changing the airfoil shape to improve the performance. This may be done in several ways:

By hand:

Using knowledge of the effects of geometry changes on C_p and C_p changes on performance.

By numerical optimization:

Using shape functions to represent the airfoil geometry and letting the computer decide on the sequence of modifications needed to improve the design. The airfoils used in this research are Gottingen airfoils which have non-dimensional linear chord length having profile thickness of unit length. Hence, the airfoils are then scaled to the chord length of the airfoil at each section. Different airfoils are noted and tested for the wind patterns and the airfoil most suited for the implementation of the wind turbine is selected.

6. GOTTINGEN PROFILES

6.1. GOTTINGEN 364 AIRFOIL

Gottingen 364 is proposed for analysis. It has Max thickness of 10.8% at 30% chord and Max camber of 6.5% at 30% chord. Table 3.2 represents the non-dimensional linear chord length and profile thickness of Gottingen 364. This profile is an asymmetric profile and profile thickness on either side of the mean line may vary based on the profile.

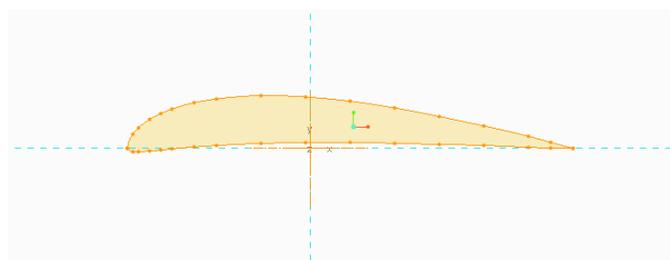


Figure 6.1. Gottingen 364 airfoil

Points	X	y1	y2
1	0	0.102	0.102
2	0.15875	0.486	0
3	0.3175	0.654	0.006
4	0.635	0.876	0.042
5	0.9525	1.032	0.066
6	12.7	1.158	0.078
7	19.05	1.32	0.126
8	25.4	1.422	0.156
9	38.1	1.5	0.204
10	50.8	1.452	0.222
11	63.5	1.332	0.216
12	76.2	1.14	0.186
13	88.9	0.906	0.15
14	101.6	0.642	0.108
15	114.3	0.348	0.054
16	120.65	0.186	0.024
17	127	0.012	0.012

Figure 6.2. Gottingen 364 non-dimensional linear chord length and profile thickness

6.2. GOTTINGEN 384 AIRFOIL

Gottingen 384 is proposed for analysis. It has Max thickness of 19.6% at 29.3% chord and Max camber of 6.9% at 39.3% chord. Table represents the non-dimensional linear chord length and profile thickness of Gottingen 384. This profile is an asymmetric profile and profile thickness on either side of the mean line may vary based on the profile.

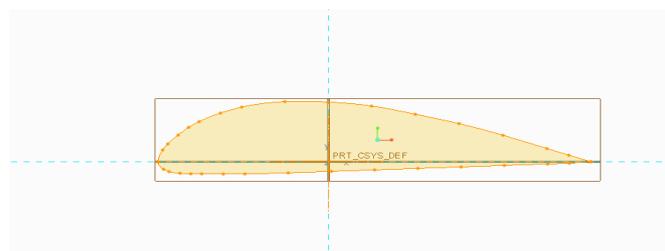


Figure 6.3. Gottingen 384 airfoil

Points	x	y1	y2
1	0	0.498	0.498
2	0.15875	0.87	0.27
3	0.3175	1.074	0.186
4	0.635	1.374	0.132
5	0.9525	1.608	0.096
6	12.7	1.794	0.066
7	19.05	2.058	0.036
8	25.4	2.226	0.018
9	38.1	2.3004	0
10	50.8	2.298	0
11	63.5	2.106	0
12	76.2	1.794	0
13	88.9	1.416	0
14	101.6	0.966	0
15	114.3	0.498	0
16	120.65	0.258	0
17	127	0	0

Figure 6.4. Gottingen 364 non-dimensional linear chord length and profile thickness

6.3. GOTTINGEN 428

Gottingen 428 is proposed for analysis. It has Max thickness of 7.9% at 30% chord and Max camber of 5.2% at 40% chord. Table represents the non-dimensional linear chord length and profile thickness of Gottingen 428. This profile is an asymmetric profile and profile thickness on either side of the mean line may vary based on the profile.

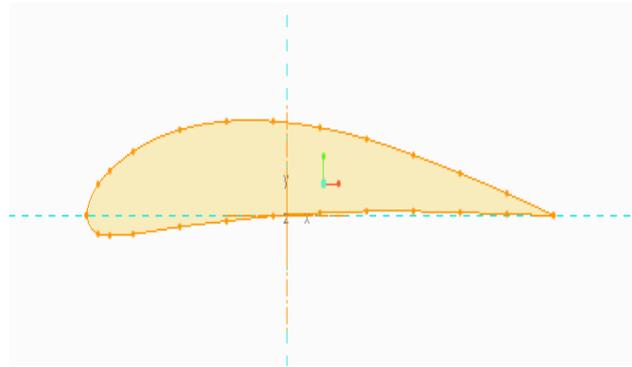


Figure 6.5. Gottingen 428 airfoil

Points	x	y1	y2
1	0	0.3	0.306
2	0.15875	0.612	0.096
3	0.3175	0.738	0.036
4	0.635	0.918	0.006
5	0.9525	10620	0
6	12.7	1.176	0.012
7	19.05	1.35	0.054
8	25.4	1.452	0.084
9	38.1	1.542	0.132
10	50.8	1.512	0.174
11	63.5	1.392	0.186
12	76.2	1.2	0.18
13	88.9	0.942	0.15
14	101.6	0.654	0.102
15	114.3	3420	0.048
16	120.65	0.174	0.024
17	127	0	0

Figure 6.6. Gottingen 428 non-dimensional linear chord length and profile thickness

6.4. GOTTINGEN 682

Gottingen 682 is proposed for analysis. It has Max thickness of 10.7% at 30% chord and Max camber of 4.3% at 40% chord. Table represents the non-dimensional linear chord length and profile thickness of Gottingen 682. This profile is an asymmetric profile and profile thickness on either side of the mean line may vary based on the profile.

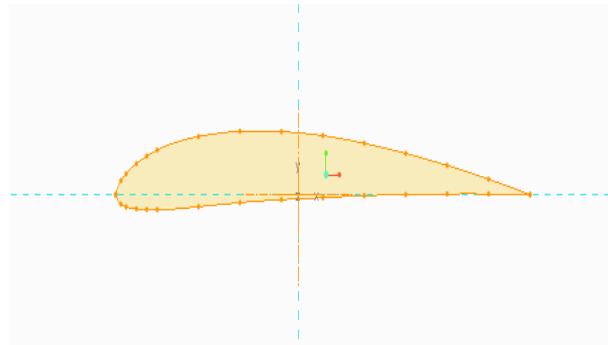


Figure 6.7. Gottingen 682 airfoil

Points	X	y1	y2
1	0	0.3	0.3
2	0.15875	0.54	0.126
3	0.3175	0.66	0.072
4	0.635	0.96	0.03
5	0.9525	0.966	0.012
6	12.7	1.068	0
7	19.05	1.2	0.06
8	25.4	1.278	0.024
Points	X	y1	y2
9	38.1	1.344	0.06
10	50.8	1.308	0.09
11	63.5	1.206	0.096
12	76.2	1.038	0.102
13	88.9	0.828	0.09
14	101.6	0.582	0.072
15	114.3	0.306	0.042
16	120.65	0.162	0.018
17	127	0	0

Figure 6.8. Gottingen 682 non-dimensional linear chord length and profile thickness

6.5. GOTTINGEN 480

Gottingen 480 is proposed for analysis. It has Max thickness of 11.8% at 30% chord and Max camber of 5.5% at 40% chord. Table represents the non-dimensional linear chord length and profile thickness of Gottingen 480. This profile is an asymmetric profile and profile thickness on either side of the mean line may vary based on the profile.

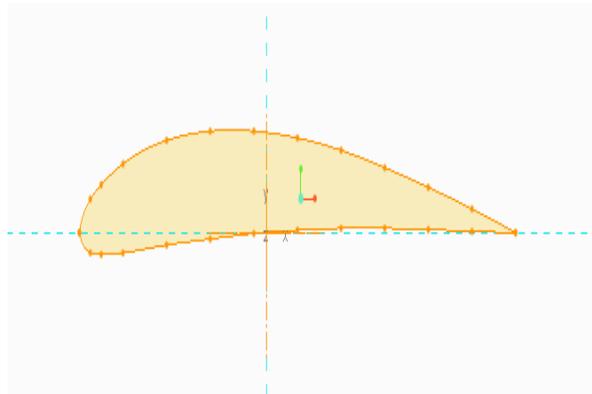


Figure 6.9. Gottingen 480 airfoil

Points	x	y1	y2
1	0	0.3	0.306
2	0.15875	0.612	0.096
3	0.3175	0.738	0.036
4	0.635	0.918	0.006
5	0.9525	10620	0
6	12.7	1.176	0.012
7	19.05	1.35	0.054
8	25.4	1.452	0.084
9	38.1	1.542	0.132
10	50.8	1.512	0.174
11	63.5	1.392	0.186
12	76.2	1.2	0.18
13	88.9	0.942	0.15
14	101.6	0.654	0.102
15	114.3	3420	0.048
16	120.65	0.174	0.024
17	127	0	0

Figure 6.10. Gottingen 480 non-dimensional linear chord length and profile thickness

7. BLADE CONSTRUCTION

For the airfoils selected, the blade construction points are calculated at different sections in the airfoil based on the chord length at each section. The blade construction table for the various airfoils are calculated and tabulated.

S.No	section 1			section 2			section 3			section 4			section 5		
	R=281.01	y _{max} =12		R=663.4	y _{max} =10		R=894.01	y _{max} =8		R=938.62	y _{max} =5.5		R=1081.4	y _{max} =3	
	l=127	β=58.02°		l=231	β=17.94°		l=187	β=11.1°		l=150	β=6.98°		l=130	β=3.83°	
	x	y1	y2	x	y1	y2	X	y1	y2	x	y1	y2	x	y1	y2
1	0	1.224	1.224	0	0.85	0.85	0	0.544	0.544	0	0.257125	0.257125	0	0.3645	0
2	0.15875	5.832	0	0.28875	4.05	0	0.23375	2.592	0	0.1875	1.225125	0	0.1625	0.4905	0.0045
3	0.3175	7.848	0.072	0.5775	5.45	0.05	0.4675	3.488	0.032	0.375	1.648625	0.015125	0.325	0.657	0.0315
4	0.635	10.512	0.504	1.155	7.3	0.35	0.935	4.672	0.224	0.75	2.20825	0.105875	0.65	0.774	0.0495
5	0.9525	12.384	0.792	1.7325	8.6	0.55	1.4025	5.504	0.352	1.125	2.6015	0.166375	0.975	0.8685	0.0585
6	12.7	13.896	0.936	23.1	9.65	0.65	18.7	6.176	0.416	15	2.919125	0.196625	13	0.99	0.0945
7	19.05	15.84	1.512	34.65	11	1.05	28.05	7.04	0.672	22.5	3.3275	0.317625	19.5	1.0665	0.117
8	25.4	17.064	1.872	46.2	11.85	1.3	37.4	7.584	0.832	30	3.584625	0.39325	26	1.125	0.153
9	38.1	18	2.448	69.3	12.5	1.7	56.1	8	1.088	45	3.78125	0.51425	39	1.089	0.1665
10	50.8	17.424	2.664	92.4	12.1	1.85	74.8	7.744	1.184	60	3.66025	0.559625	52	0.999	0.162
11	63.5	15.984	2.592	115.5	11.1	1.8	93.5	7.104	1.152	75	3.35775	0.5445	65	0.855	0.1395
12	76.2	13.68	2.232	138.6	9.5	1.55	112.2	6.08	0.992	90	2.87375	0.468875	78	0.6795	0.1125
13	88.9	10.872	1.8	161.7	7.55	1.25	130.9	4.832	0.8	105	2.283875	0.378125	91	0.4815	0.081
14	101.6	7.704	1.296	184.8	5.35	0.9	149.6	3.424	0.576	120	1.618375	0.27225	104	0.261	0.0405
15	114.3	4.176	0.648	207.9	2.9	0.45	168.3	1.856	0.288	135	0.87725	0.136125	117	0.1395	0.018
16	120.65	2.232	0.288	219.45	1.55	0.2	177.65	0.992	0.128	142.5	0.468875	0.0605	123.5	0.009	0.009
17	127	0.144	0.144	231	0.1	0.1	187	0.064	0.064	150	0.03025	0.03025	130	0.0765	0.0765

Figure 7.1. Blade construction table Gottingen 364

Sno	section 1			section 2			section 3			section 4			section 5		
	R=281.01	y _{max} =12		R=663.4	y _{max} =10		R=894.01	y _{max} =8		R=938.62	y _{max} =5.5		R=1081.4	y _{max} =3	
	l=127	β=58.02°		l=231	β=17.94°		l=187	β=11.1°		l=150	β=6.98°		l=130	β=3.83°	
	x	y1	y2	x	y1	y2	X	y1	y2	x	y1	y2	x	y1	y2
1	0	5.976	5.976	0	4.15	4.15	0	2.656	2.656	0	1.255375	1.255375	0	0.3735	0.3735
2	0.15875	10.44	3.24	0.28875	7.25	2.25	0.23375	4.64	1.44	0.1875	2.193125	0.680625	0.1625	0.6525	0.2025
3	0.3175	12.888	2.232	0.5775	8.95	1.55	0.4675	5.728	0.992	0.375	2.707375	0.468875	0.325	0.8055	0.1395
4	0.635	16.488	1.584	1.155	11.45	1.1	0.935	7.328	0.704	0.75	3.463625	0.33275	0.65	1.0305	0.099
5	0.9525	19.296	1.152	1.7325	13.4	0.8	1.4025	8.576	0.512	1.125	4.0535	0.242	0.975	1.206	0.072
6	12.7	21.528	0.792	23.1	14.95	0.55	18.7	9.568	0.352	15	4.522375	0.166375	13	1.3455	0.0495
7	19.05	24.696	0.432	34.65	17.15	0.3	28.05	10.976	0.192	22.5	5.187875	0.09075	19.5	1.5435	0.027
8	25.4	26.712	0.216	46.2	18.55	0.15	37.4	11.872	0.096	30	5.611375	0.045375	26	1.6695	0.0135
9	38.1	27.6048	0	69.3	19.17	0	56.1	12.2688	0	45	5.798925	0	39	1.7253	0
10	50.8	27.576	0	92.4	19.15	0	74.8	12.256	0	60	5.792875	0	52	1.7235	0
11	63.5	25.272	0	115.5	17.55	0	93.5	11.232	0	75	5.308875	0	65	1.5795	0
12	76.2	21.528	0	138.6	14.95	0	112.2	9.568	0	90	4.522375	0	78	1.3455	0
13	88.9	16.992	0	161.7	11.8	0	130.9	7.552	0	105	3.5695	0	91	1.062	0
14	101.6	11.592	0	184.8	8.05	0	149.6	5.152	0	120	2.435125	0	104	0.7245	0
15	114.3	5.976	0	207.9	4.15	0	168.3	2.656	0	135	1.255375	0	117	0.3735	0
16	120.65	3.096	0	219.45	2.15	0	177.65	1.376	0	142.5	0.650375	0	123.5	0.1935	0
17	127	0	0	231	0	0	187	0	0	150	0	0	130	0	0

Figure 7.2. Blade construction table Gottingen 384

Sno	section 1			section 2			section 3			section 4			section 5			
	R=281.01		ymax=12	R=663.4		ymax=10	R=894.01		ymax=8	R=938.62		ymax=5.5		R=1081.4		ymax=3
	l=127	$\beta=58.02^\circ$		l=231	$\beta=17.94^\circ$		l=187	$\beta=11.1^\circ$		l=150	$\beta=6.98^\circ$		l=130	$\beta=3.83^\circ$		
	x	y1	y2	x	y1	y2	X	y1	y2	x	y1	y2	x	y1	y2	
1	0	0.15	0.15	0	0.125	0.125	0	0.1	0.1	0	0.06875	0.06875	0	0.0375	0.0375	
2	0.15875	0.33	0.036	0.28875	0.275	0.03	0.23375	0.22	0.024	0.1875	0.15125	0.0165	0.1625	0.0825	0.009	
3	0.3175	0.42	0.024	0.5775	0.35	0.02	0.4675	0.28	0.016	0.375	0.1925	0.011	0.325	0.105	0.006	
4	0.635	0.576	0.012	1.155	0.48	0.01	0.935	0.384	0.008	0.75	0.264	0.0055	0.65	0.144	0.003	
5	0.9525	0.726	0	1.7325	0.605	0	1.4025	0.484	0	1.125	0.33275	0	0.975	0.1815	0	
6	12.7	0.78	0	23.1	0.65	0	18.7	0.52	0	15	0.3575	0	13	0.195	0	
7	19.05	0.906	0.006	34.65	0.755	0.005	28.05	0.604	0.004	22.5	0.41525	0.00275	19.5	0.2265	0.0015	
8	25.4	0.984	0.018	46.2	0.82	0.015	37.4	0.656	0.012	30	0.451	0.00825	26	0.246	0.0045	
9	38.1	1.026	0.036	69.3	0.855	0.03	56.1	0.684	0.024	45	0.47025	0.0165	39	0.2565	0.009	
10	50.8	1.002	0.048	92.4	0.835	0.04	74.8	0.668	0.032	60	0.45925	0.022	52	0.2505	0.012	
11	63.5	0.936	0.048	115.5	0.78	0.04	93.5	0.624	0.032	75	0.429	0.022	65	0.234	0.012	
12	76.2	0.816	0.042	138.6	0.68	0.035	112.2	0.544	0.028	90	0.374	0.01925	78	0.204	0.0105	
13	88.9	0.66	0.03	161.7	0.55	0.025	130.9	0.44	0.02	105	0.3025	0.01375	91	0.165	0.0075	
14	101.6	0.504	0.018	184.8	0.42	0.015	149.6	0.336	0.012	120	0.231	0.00825	104	0.126	0.0045	
15	114.3	0.258	0.006	207.9	0.215	0.005	168.3	0.172	0.004	135	0.11825	0.00275	117	0.0645	0.0015	
16	120.65	0.144	0	219.45	0.12	0	177.65	0.096	0	142.5	0.066	0	123.5	0.036	0	
17	127	0	0	231	0	0	187	0	0	150	0	0	130	0	0	

Figure 7.3. Blade construction table Gottingen 428

Sno	section 1			section 2			section 3			section 4			section 5			
	R=281.0		ymax=12	R=663.4		ymax=10	R=894.0		ymax=8	R=938.6		ymax=5.5		R=1081.4		ymax=3
	l=127	$\beta=58.02^\circ$		l=231	$\beta=17.94^\circ$		l=187	$\beta=11.1^\circ$		l=150	$\beta=6.98^\circ$		l=130	$\beta=3.83^\circ$		
	x	y1	y2	x	y1	y2	X	y1	y2	x	y1	y2	x	y1	y2	
1	0	1.8	1.8	0	1.25	1.25	0	0.8	0.8	0	0.378125	0.378125	0	0.1125	0.1125	
2	0.15875	3.96	0.432	0.28875	2.75	0.3	0.23375	1.76	0.192	0.1875	0.831875	0.09075	0.1625	0.2475	0.027	
3	0.3175	5.04	0.288	0.5775	3.5	0.2	0.4675	2.24	0.128	0.375	1.05875	0.0605	0.325	0.315	0.018	
4	0.635	6.912	0.144	1.155	4.8	0.1	0.935	3.072	0.064	0.75	1.452	0.03025	0.65	0.432	0.009	
5	0.9525	8.712	0	1.7325	6.05	0	1.4025	3.872	0	1.125	1.830125	0	0.975	0.5445	0	
6	12.7	9.36	0	23.1	6.5	0	18.7	4.16	0	15	1.96625	0	13	0.585	0	
7	19.05	10.872	0.072	34.65	7.55	0.05	28.05	4.832	0.032	22.5	2.283875	0.015125	19.5	0.6795	0.0045	
8	25.4	11.808	0.216	46.2	8.2	0.15	37.4	5.248	0.096	30	2.4805	0.045375	26	0.738	0.0135	
9	38.1	12.312	0.432	69.3	8.55	0.3	56.1	5.472	0.192	45	2.586375	0.09075	39	0.7695	0.027	
10	50.8	12.024	0.576	92.4	8.35	0.4	74.8	5.344	0.256	60	2.525875	0.121	52	0.7515	0.036	
11	63.5	11.232	0.576	115.5	7.8	0.4	93.5	4.992	0.256	75	2.3595	0.121	65	0.702	0.036	
12	76.2	9.792	0.504	138.6	6.8	0.35	112.2	4.352	0.224	90	2.057	0.105875	78	0.612	0.0315	
13	88.9	7.92	0.36	161.7	5.5	0.25	130.9	3.52	0.16	105	1.66375	0.075625	91	0.495	0.0225	
14	101.6	6.048	0.216	184.8	4.2	0.15	149.6	2.688	0.096	120	1.2705	0.045375	104	0.378	0.0135	
15	114.3	3.096	0.072	207.9	2.15	0.05	168.3	1.376	0.032	135	0.650375	0.015125	117	0.1935	0.0045	
16	120.65	1.728	0	219.45	1.2	0	177.65	0.768	0	142.5	0.363	0	123.5	0.108	0	
17	127	0	0	231	0	0	187	0	0	150	0	0	130	0	0	

Figure 7.4. Blade construction table Gottingen 480

Sno	section 1			section 2			section 3			section 4			section 5			
	R=281.01		ymax=12	R=663.4		ymax=10	R=894.01		ymax=8	R=938.62		ymax=5.5		R=1081.4		ymax=3
	l=127		$\beta=58.02^\circ$	l=231		$\beta=17.94^\circ$	l=187		$\beta=11.1^\circ$	l=150		$\beta=6.98^\circ$		l=130		$\beta=3.83^\circ$
	X	y1	y2	X	y1	y2	X	y1	y2	x	y1	y2	x	y1	y2	
1	0	3.6	3.6	0	2.5	2.5	0	1.6	1.6	0	0.75625	0.75625	0	0.225	0.225	
2	0.15875	6.48	1.512	0.28875	4.5	1.05	0.23375	2.88	0.672	0.1875	1.36125	0.317625	0.1625	0.405	0.0945	
3	0.3175	7.92	0.864	0.5775	5.5	0.6	0.4675	3.52	0.384	0.375	1.66375	0.1815	0.325	0.495	0.054	
4	0.635	11.52	0.36	1.155	8	0.25	0.935	5.12	0.16	0.75	2.42	0.075625	0.65	0.72	0.0225	
5	0.9525	11.592	0.144	1.7325	8.1	0.1	1.4025	5.152	0.064	1.125	2.435125	0.03025	0.975	0.7245	0.009	
6	12.7	12.816	0	23.1	8.9	0	18.7	5.696	0	15	2.69225	0	13	0.801	0	
7	19.05	14.4	0.72	34.65	10	0.5	28.05	6.4	0.32	22.5	3.025	0.15125	19.5	0.9	0.045	
8	25.4	15.336	0.288	46.2	10.7	0.2	37.4	6.816	0.128	30	3.221625	0.0605	26	0.9585	0.018	
9	38.1	16.128	0.72	69.3	11.2	0.5	56.1	7.168	0.32	45	3.388	0.15125	39	1.008	0.045	
10	50.8	15.696	1.08	92.4	10.9	0.75	74.8	6.976	0.48	60	3.29725	0.226875	52	0.981	0.0675	
11	63.5	14.472	1.152	115.5	10.1	0.8	93.5	6.432	0.512	75	3.040125	0.242	65	0.9045	0.072	
12	76.2	12.456	1.224	138.6	8.7	0.85	112.2	5.536	0.544	90	2.616625	0.257125	78	0.7785	0.0765	
13	88.9	9.936	1.08	161.7	6.9	0.75	130.9	4.416	0.48	105	2.08725	0.226875	91	0.621	0.0675	
14	101.6	6.984	0.864	184.8	4.9	0.6	149.6	3.104	0.384	120	1.467125	0.1815	104	0.4365	0.054	
15	114.3	3.672	0.504	207.9	2.6	0.35	168.3	1.632	0.224	135	0.771375	0.105875	117	0.2295	0.0315	
16	120.65	1.944	0.216	219.45	1.4	0.15	177.65	0.864	0.096	142.5	0.408375	0.045375	123.5	0.1215	0.0135	
17	127	0	0	231	0	0	187	0	0	150	0	0	130	0	0	

Figure 7.5. Blade construction table Gottingen682

8. NUMERICAL METHOD

8.1. NUMERICAL ANALYSIS

- The modelled blade is saved in the .stp (STEP) format.
- This model is imported into ANSYS 14.5 work bench in the FLUENT solver.
- In the design modeler of the ANSYS the fluid domain is modelled with a inlet pipe, MRF zone and the outlet pipe.
- The blade is positioned in the MRF zone. This is then imported into the mesh module of the analysis software.
- A patch conforming mesh is done.
- The mesh is now imported to the fluent solver and the necessary boundary conditions are defined.
- The simulation is run till convergence of the answer.

The results obtained are plotted and compared with the theoretical results.

8.2. ANALYSIS PROCEDURE

- Building of blades
- For modelling the turbine, we first select the airfoil section. Based on the information from the blade construction table, the airfoil sections of various chord lengths are placed at specific sections in the blade.
- The airfoil points are imported in to the modeler and a surface is created as shown in Figure along the various airfoil sections as shown in Figure.

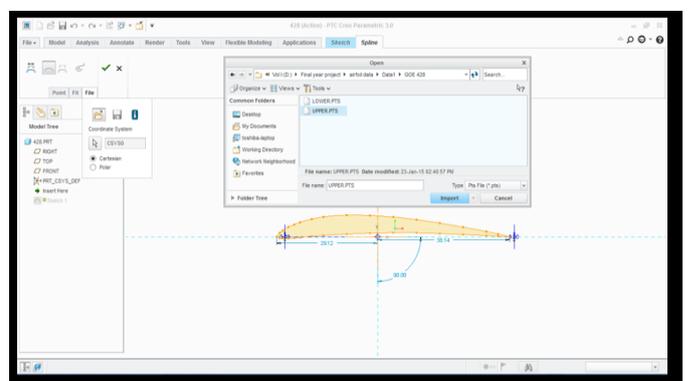


Figure 8.1. AIRFOIL IMPORT FOR BLADE CONSTRUCTION

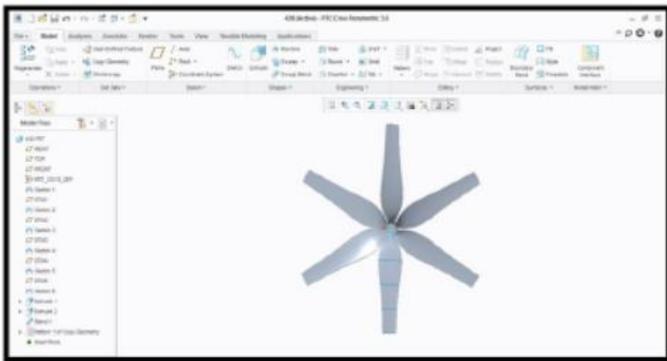


Figure 8.2. MODELLED BLADE

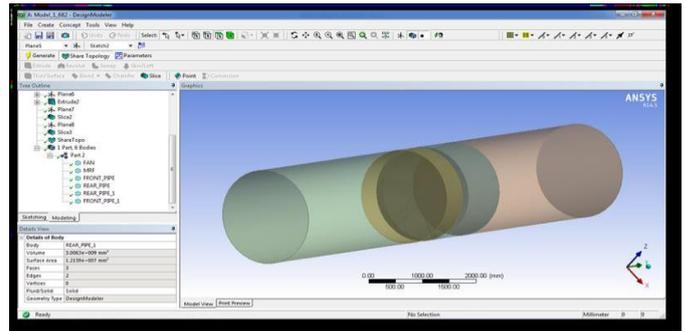


Figure 8.4. Analysis model with MRF model along with the inlet and outlet

8.3. FLUID DOMAIN GENERATION

- The modelled wind turbine is imported into the CFD commercial software(ANSYS-Workbench).
- The model is now imported into the geometry component of the solver as shown in Figure .
- Now open the design modeller and generate the geometry.
- Choose on which plane to add the geometry.
- After import a green tick should accompany the import file as shown in the image below.
- Now we create an enclosure around the turbine and name it the moving reference zone (MRF).
- The inlet pipe is now modelled from the MRF zone.
- The similar procedure is carried out for the outlet pipe.
- In order to increase the accuracy of the analysis the inlet and outlet pipes are sliced at some distance from the MRF frame so that the flow around the blade is consistent as displayed.
- Fully constructed wind turbine model is ready to be meshed.

- The model is saved and then updated into the mesh component as shown in Figure .
- Once the model is updated in the mesh, a patch conforming mesh as shown in Figure is done and a suitable body sizing is given. The mesh around the MRF zone should be finer compared to the rest of the fluid domain so that an accurate result can be obtained.

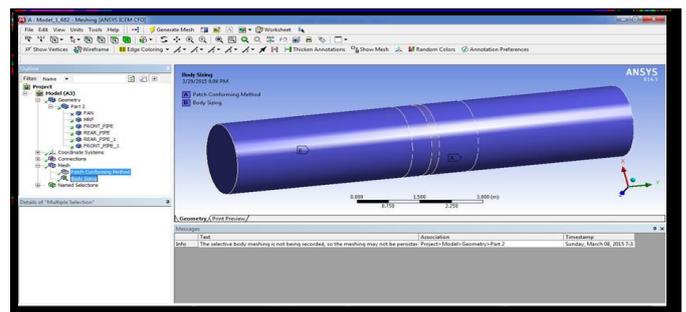


Figure.8.5. Patch Conforming meshing

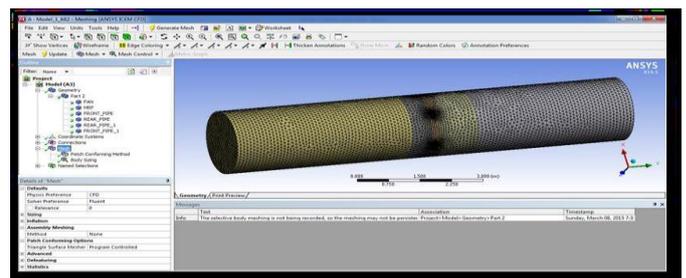


Figure 8.6. Figure Meshed model

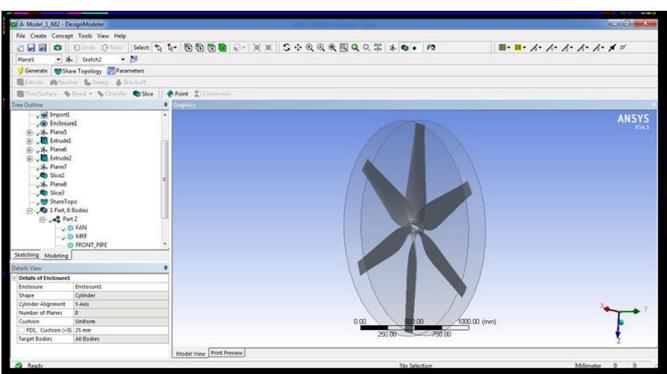


Figure 8.3. Blade with MRF

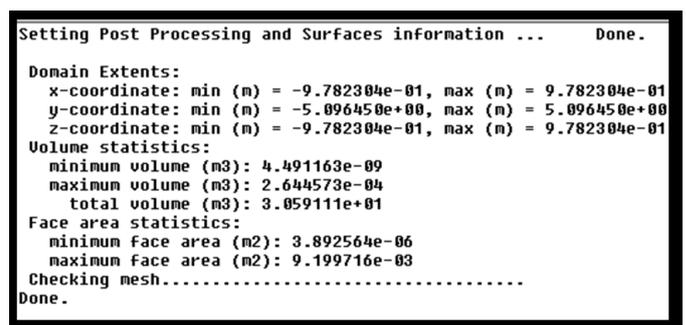


Figure 8.7. Mesh check

- The mesh is saved and the fluent setup is launched. Since the system we are using is a 4 core system, we input the number of parallel processes as 4 and choose the double precision method as shown in Figure .
- Once the model is loaded in the solver, a mesh check is done to ensure that the model is meshed properly. The scale of the model is also checked as displayed in Figure .
- Similarly, for all the wind turbine models which is designed with Gottingen airfoils the patch confirming mesh is provided and the details is given table below.

MODEL	GOE364	GOE384	GOE 482	GOE480	GOE682
MESH TYPE	Triangular mesh				
MESH COUNT	1500000	1500000	1500000	1500000	1500000
SKEWNESS	0.7	0.7	0.7	0.7	0.7

- The solver type is chosen to be pressure based and the time function as steady state.
- Due to lack of computational power we cannot do a transient state analysis due to the huge volume of data to be processed.
- The domain consists an inlet zone and outlet zone and MRF zone as shown in the above figure.
- For Inlet the velocity inlet is opted & for outlet pressure outlet is taken.
- The Realizable K-ε Model is chosen along with a standard wall function.
- The cell zone condition of the MRF is given a frame motion and made to rotate at -300 RPM (counter clockwise rotation).
- The analysis done here is an inverse analysis where the turbine is rotated at 300 RPM and the corresponding inlet velocity due to the suction created by it is recorded.
- Hence based on the airfoil profile used, the inlet velocity will vary.
- This can then be used to match the profile with the place.
- This will prove that the corresponding inlet velocity which is to be obtained only for that profile.
- The solution is initialized and the calculation is run till it converges.
- The procedure stated above is repeated for various blades and the converged solution is obtained.
- The results are then obtained from the CFD post processor.

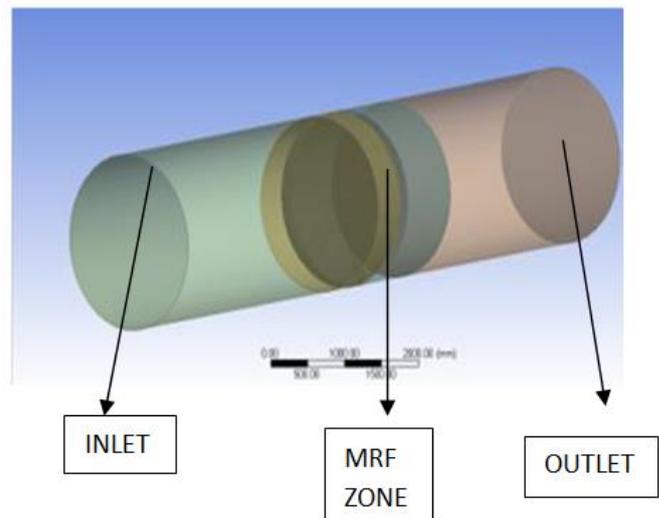


Figure 8.8. Boundary conditions with inlet ,outlet and MRF

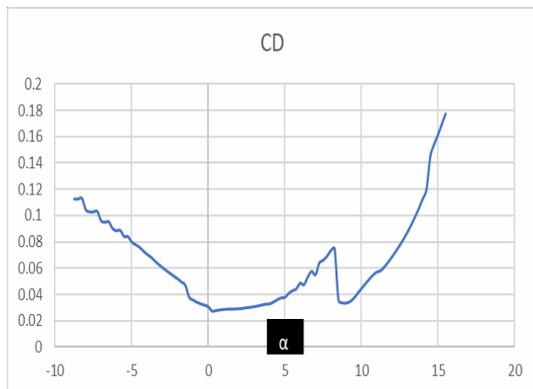
- The domain consists an inlet zone and outlet zone and MRF zone as shown in the above figure.
- For Inlet the velocity inlet is opted & for outlet pressure outlet is taken.
- The Realizable K-ε Model is chosen along with a standard wall function.
- The cell zone condition of the MRF is given a frame motion and made to rotate at -300 RPM (counter clockwise rotation).
- The analysis done here is an inverse analysis where the turbine is rotated at 300 RPM and the corresponding inlet velocity due to the suction created by it is recorded.
- Hence based on the airfoil profile used, the inlet velocity will vary.
- This can then be used to match the profile with the place.
- This will prove that the corresponding inlet velocity which is to be obtained only for that profile.
- The solution is initialized and the calculation is run till it converges.
- The procedure stated above is repeated for various blades and the converged solution is obtained.
- The results are then obtained from the CFD post processor.

9. NUMERICAL RESULTS

The computational results for the different airfoils for the given blade are discussed. The performance of the airfoils are also displayed in this chapter. After the solution is converged, the CFD post processing method is used to obtain the various results are shown in form of graph and contours. The various results obtained from the analysis are

- Coefficient of Lift

- Coefficient of drag
- Streamlines
- Velocity contour
- Pressure contour



9.1. WIND TURBINE ROTOR GOE-364

Figure 9.1 GOE-364 CD vs α (Angle of attack)

The above graph shown represents CD vs α (Angle of attack) where maximum CD of the airfoil reaches at 15 Degree of angle of attack.

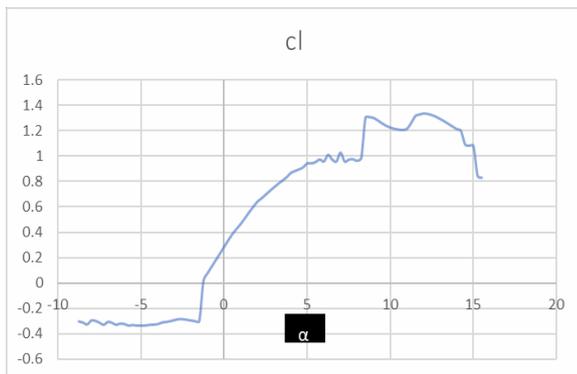


Figure 9.2. GOE 364 CL vs α (Angle of attack)

The above graph shown represents Cl vs α (Angle of attack) where maximum Cl of the airfoil reaches at 9 Degree of Angle of attack and Cl value is 1.3 respectively.

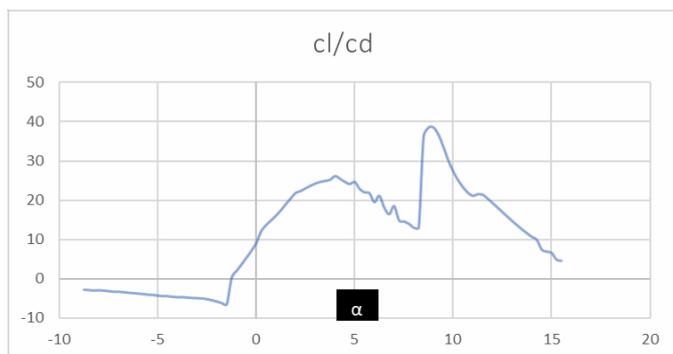


Figure 9.3. GOE 364 cl/cd vs α (Angle of attack)

The above graph shown represents cl/cd vs α (Angle of attack) where maximum cl/cd of the airfoil reaches at 9 Degree of Angle of attack and cl/cd value is 40 respectively.

9.1.1. CONTOURS OF STREAMLINE

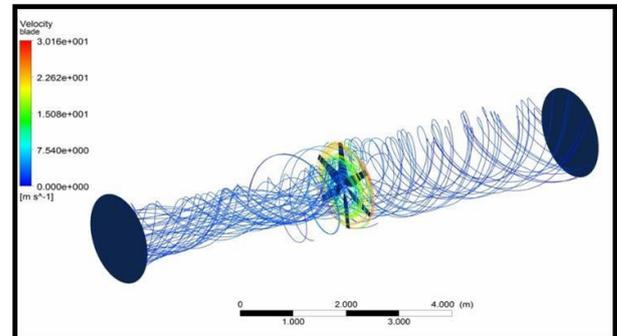


Figure 9.4. GOE 364 Streamlines

The above explains the flow of the wind as it hits the blade. This flow varies every second and the streamlines of air passes through the wind turbine. We can see that the wind flow is concentrated towards the hub. As the turbine is rotating the air in front of it sucked in and pushed out. This will cause a pressure drop in front of the turbine. The blade angle is set in such a way that the starting torque is high. Owing to the rotation of the turbine at 300 RPM we get a velocity of 3.36 m/s in the inlet. Similar results can be obtained for the other blades with various airfoil sections.

9.1.2. CONTOURS OF VELOCITY MAGNITUDE

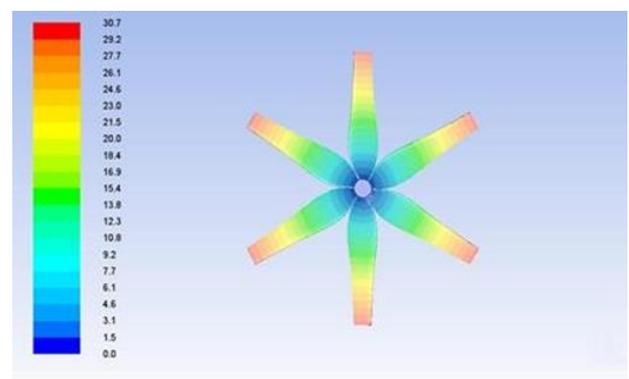


Figure 9.5 Velocity Contours- GOE 364

9.1.3. PRESSURE CONTOURS

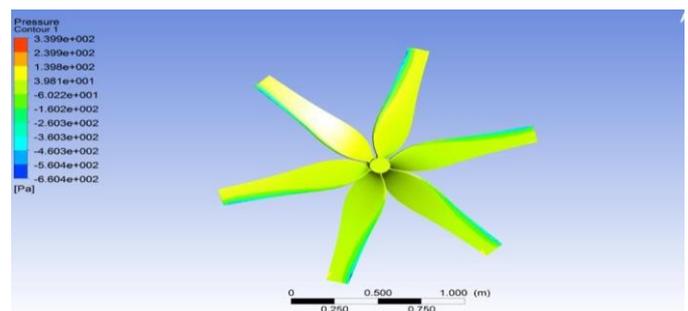


Figure 9.6 Pressure contours- GOE 364

Differential of the pressure distribution on blade surface which is obtained due rotation effect, flow separation over the blade closer to the root of blade the stronger this effect will be.

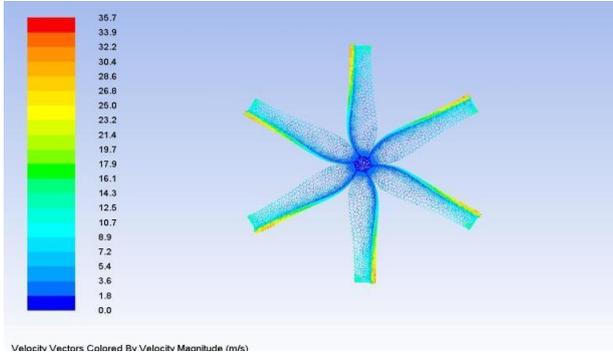


Figure 9.7. Velocity vector of wind turbine blade-GOE 364 airfoil

9.2. WIND TURBINE ROTOR GOE-384

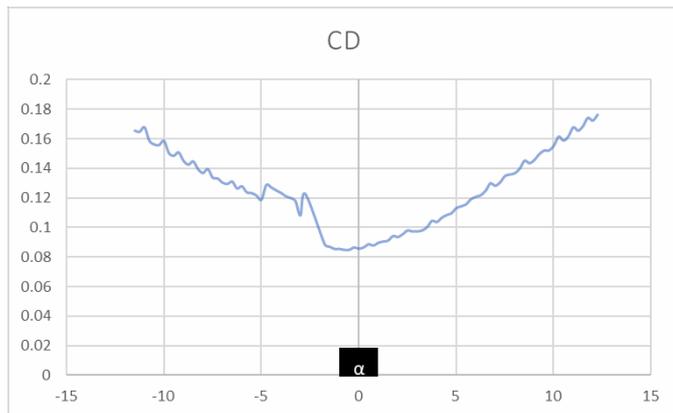


Figure 9.8. GOE-384 CD vs alpha (Angle of attack)

The above graph shown represents CD vs alpha (Angle of attack) where maximum CD of the airfoil reaches at 12.5 Degree of angle of attack.

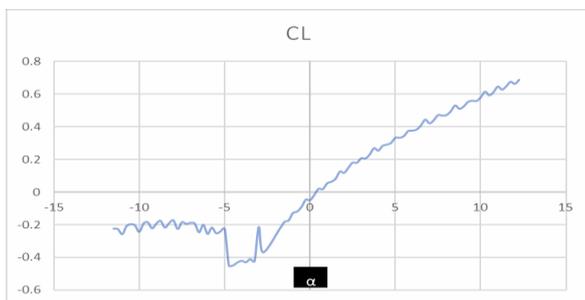


Figure 9.9 GOE 384 CL vs alpha (Angle of attack)

The above graph shown in the figure, represents Cl vs alpha (Angle of attack) where maximum Cl of the airfoil reaches at 13 Degree of Angle of attack and Cl value is 0.54 respectively.

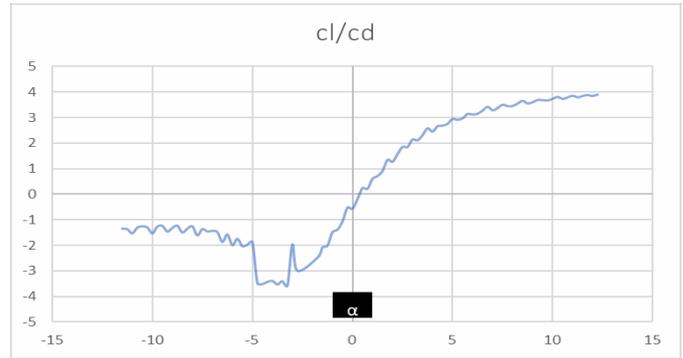


Figure 9.10 GOE 384 cl/cd vs alpha (Angle of attack)

The above graph shown represents cl/cd vs alpha (Angle of attack) where maximum cl/cd of the airfoil reaches at 9 Degree of Angle of attack and cl/cd value is 40 respectively.

9.2.1. CONTOURS OF STREAMLINE

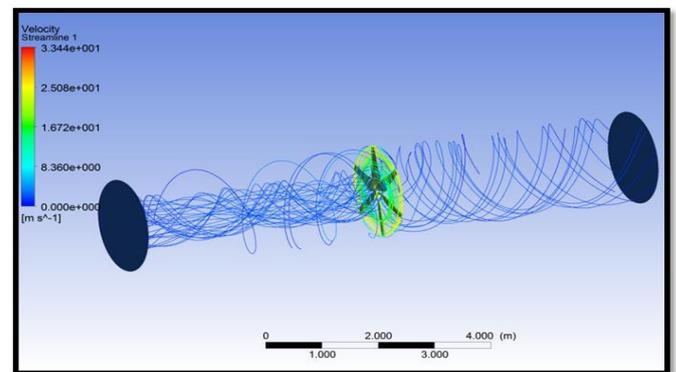


Figure 9.11. GOE 384 Streamlines

The inlet velocity obtained from this airfoil is 3.13m/s from the figure we can clearly see that the velocity will be less as compared to the 364 airfoil from the streamlines that is obtained from the analysis. The streamlines are less dense compared to the 364 airfoil.

9.2.2. CONTOUR OF VELOCITY MAGNITUDE

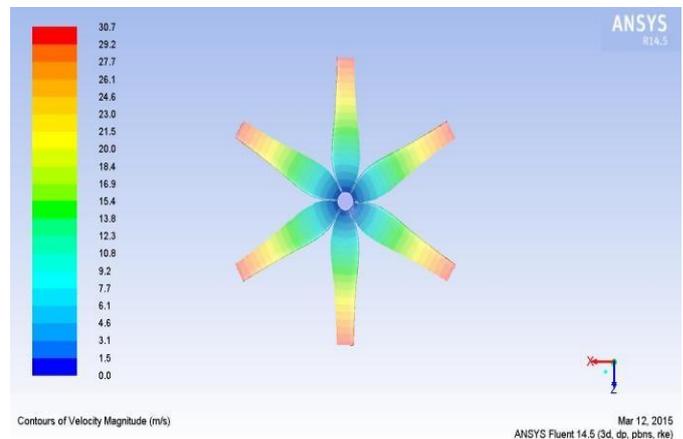


Figure 9.12 Velocity Contours- GOE 384

9.2.3. PRESSURE CONTOURS

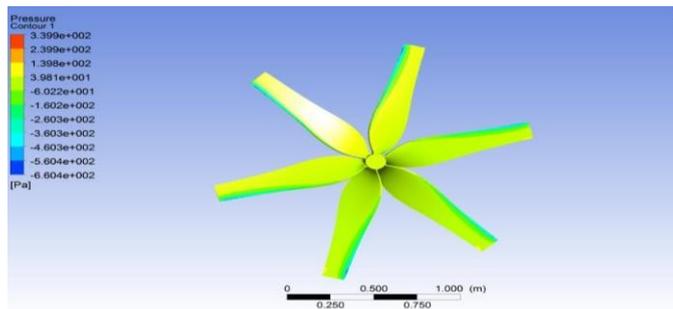


Figure 9.13 Pressure contours- GOE 384

Differential of the pressure distribution on blade surface which is obtained due to mainly from the three-dimensional rotation effect and flow separation on the blade and closer to the root of blade the stronger this effect will be.

9.2.4. VELOCITY VECTOR COLOURED BY VELOCITY MAGNITUDE

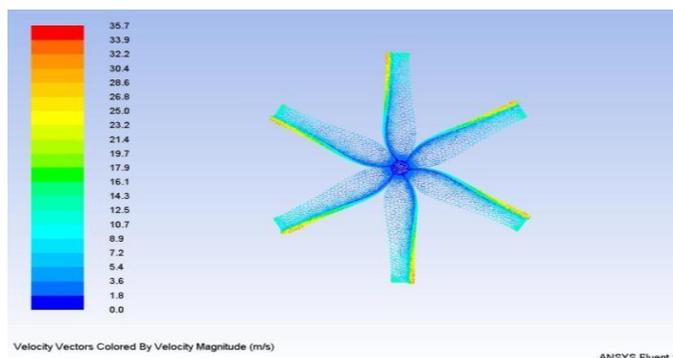


Figure 9.14 Velocity Contours- GOE 384

9.3. WIND TURBINE ROTOR SYSTEM GOE-428

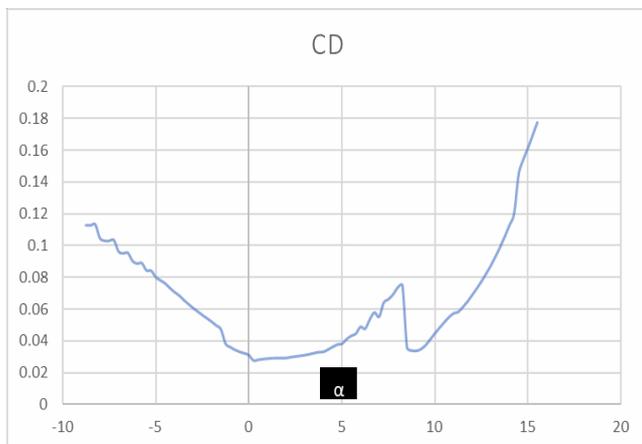


Figure 9.15. GOE-428 CD vs α (Angle of attack)

The above graph shown represents CD vs α (Angle of attack) where maximum CD of the airfoil reaches at 16 Degree of angle of attack.

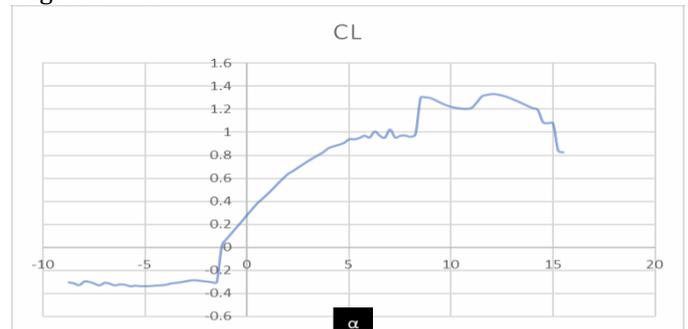


Figure 9.16 GOE 428 CL vs α (Angle of attack)

The above graph shown represents Cl vs α (Angle of attack) where maximum Cl of the airfoil reaches at 9 and at 13 Degree of Angle of attack and Cl value is 1.3 respectively.

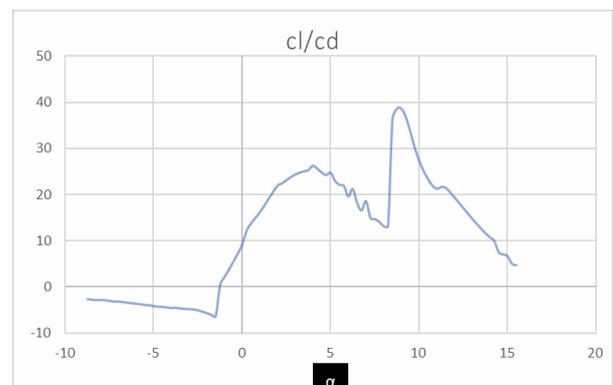


Figure 9.17 GOE 428 cl/cd vs α (Angle of attack)

The above graph shown represents cl/cd vs α (Angle of attack) where maximum cl/cd of the airfoil reaches at 9 Degree of Angle of attack and cl/cd value is 40 respectively.

9.3.1. CONTOURS OF STREAMLINE

It is clearly visible from above figure that from the streamlines the inlet velocity should be higher than that of the 364 and 384 airfoils owing to the density of the streamlines in the inlet pipe. The velocity obtained here is 3.7 m/s.

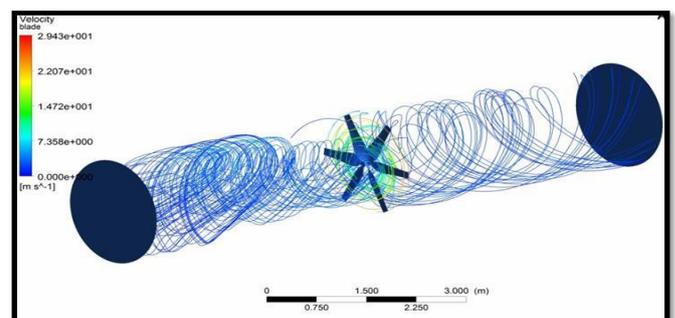


Figure 9.18 GOE 384 Streamlines

9.3.2. CONTOURS OF VELOCITY MAGNITUDE

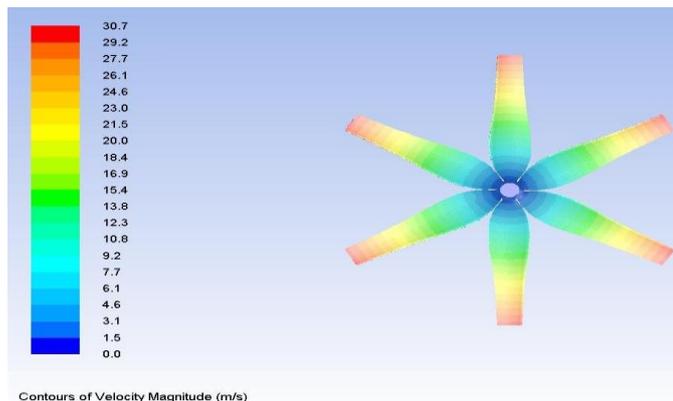


Figure 9.19 Velocity Contours- GOE 428

9.3.3. PRESSURE CONTOURS

Similar to the wind turbine rotor system of Gottingen airfoil Differential of the pressure distribution on blade surface which is obtained due to mainly from the three-dimensional rotation effect and flow separation on the blade and closer to the root of blade the stronger this effect will be. But minor changes can be viewed near the leading edge of the blade.

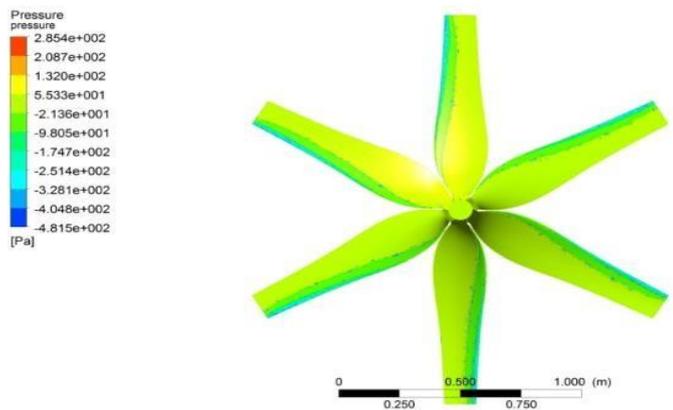


Figure 9.20 Pressure contours- GOE 428

9.3.4. VELOCITY VECTOR COLOURED BY VELOCITY

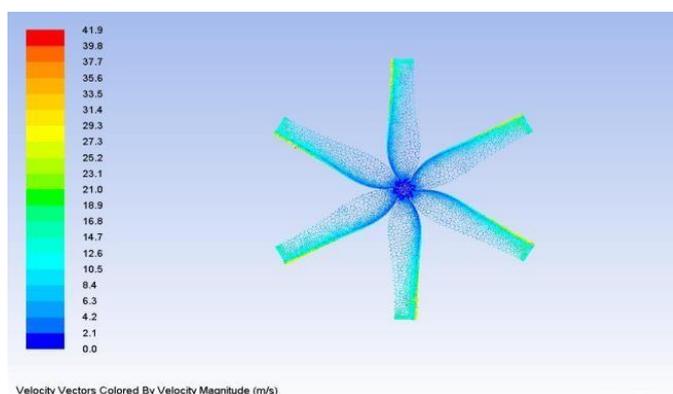


Figure 9.21 Velocity Contours- GOE 428

9.4. WIND TURBINE ROTOR SYSTEM GOE-480

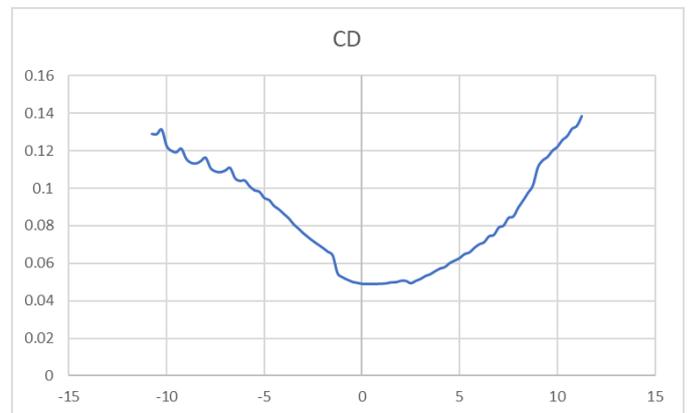


Figure 9.22 GOE-480 CD vs α (Angle of attack)

The above graph shown represents CD vs α (Angle of attack) where maximum CD of the airfoil reaches at 13 Degree of angle of attack.

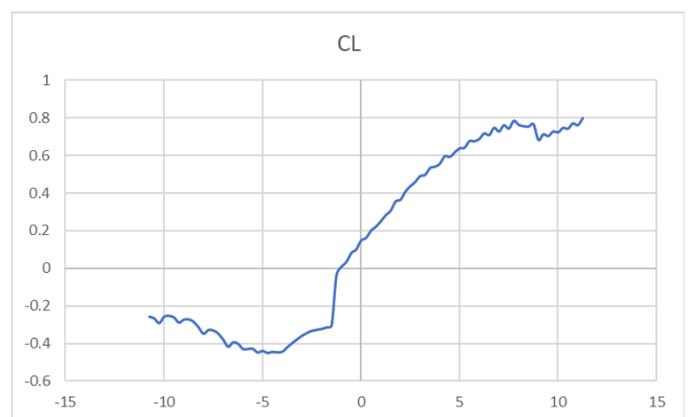


Figure 9.23 GOE 480 CL vs α (Angle of attack)

The above graph shown represents Cl vs α (Angle of attack) where maximum Cl of the airfoil reaches at 12 and Degree of Angle of attack and Cl value is 0.8 respectively.

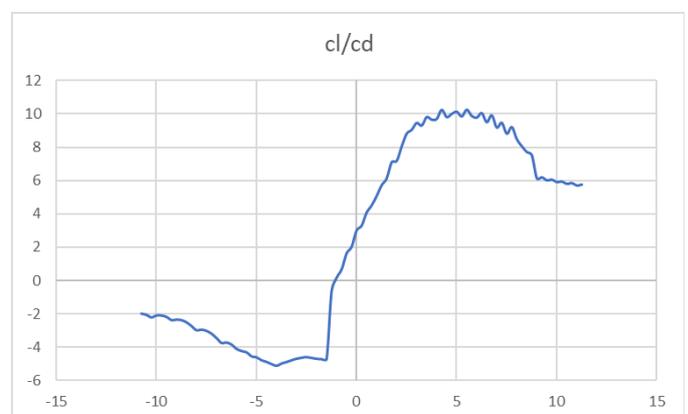


Figure 9.23 GOE 480 cl/cd vs α (Angle of attack)

The above graph shown represents c_l/c_d vs α (Angle of attack) where maximum c_l/c_d of the airfoil reaches at 5 Degree of Angle of attack and c_l/c_d value is 10 respectively.

9.4.1. CONTOURS OF STREAMLINE

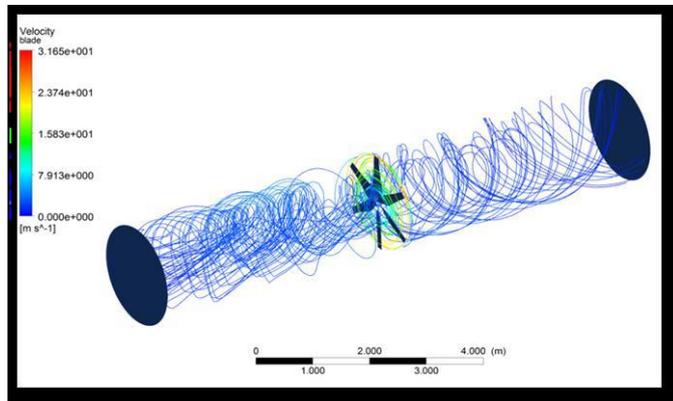


Figure 9.24 GOE 480 Streamlines

The streamlines as shown in the above figure are like that of the 428 airfoil which gives a similar inlet velocity of 3.5m/s.

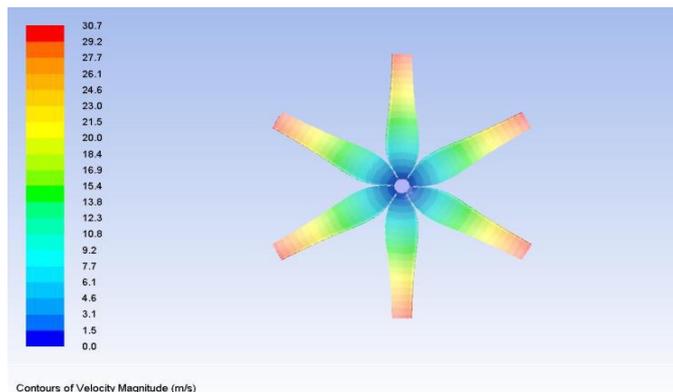


Figure 9.25 Velocity Contours- GOE 480

9.4.2. PRESSURE CONTOURS

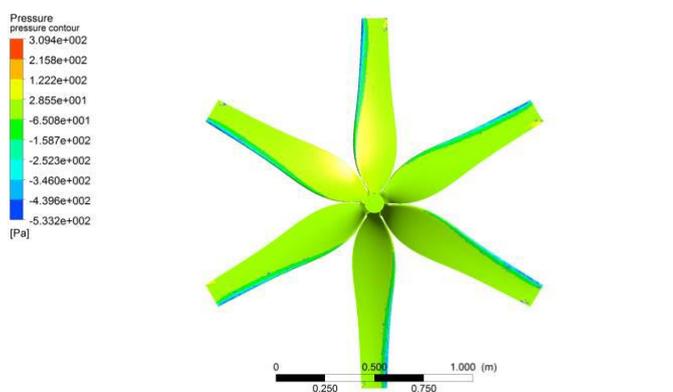


Figure 9.26 Pressure contours- GOE 480

Similar to the wind turbine rotor system of Gottingen airfoil Differential of the pressure distribution on blade surface which is obtained due to mainly from the three-dimensional rotation effect and flow separation on the blade and closer to the root of blade the stronger this effect will be. But minor changes can be viewed near the leading edge of the blade.

9.4.3. VELOCITY VECTOR COLOURED BY VELOCITY MAGNITUDE

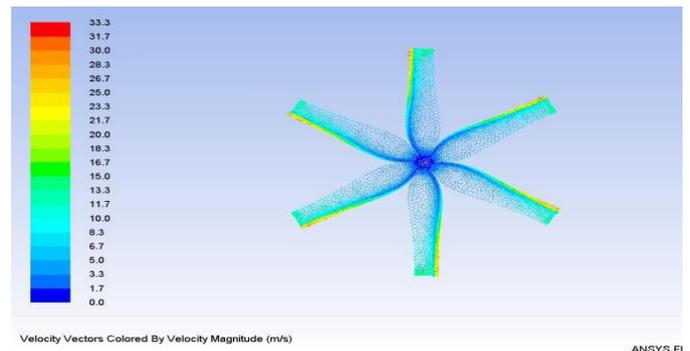


Figure 9.27 Velocity Contours- GOE 480

9.5. WIND TURBINE ROTOR SYSTEM GOE-682

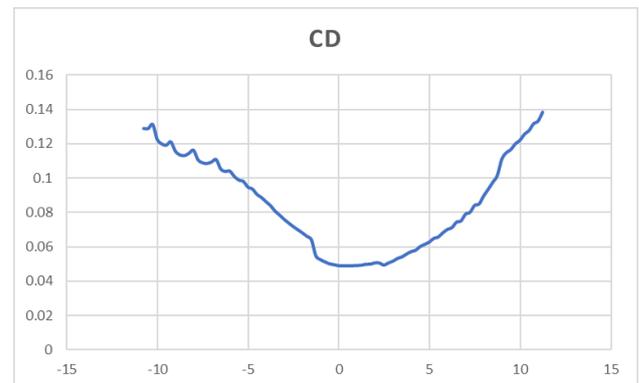


Figure 9.28 GOE-682 CD vs α (Angle of attack)

The above graph shown represents C_D vs α (Angle of attack) where maximum C_D of the airfoil reaches at 13 Degree of angle of attack.

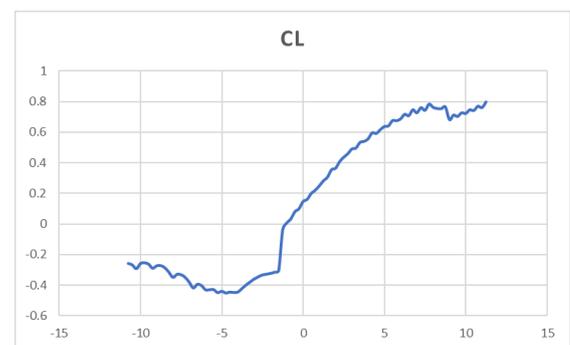


Figure 9.29 GOE 682 CL vs α (Angle of attack)

The above graph shown represents C_l vs α (Angle of attack) where maximum C_l of the airfoil reaches at 9 and at 13 Degree of Angle of attack and C_l value is 1.3 respectively.

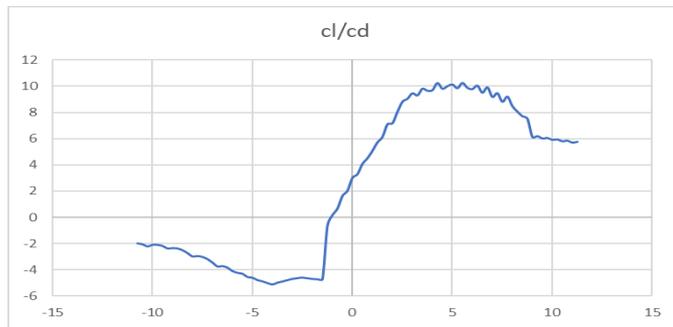


Figure 9.30 GOE 480 c_l/c_d vs α (Angle of attack)

The above graph shown represents c_l/c_d vs α (Angle of attack) where maximum c_l/c_d of the airfoil reaches at 4.7 Degree of Angle of attack and c_l/c_d value is 10 respectively.

9.5.1. CONTOURS OF STREAMLINE

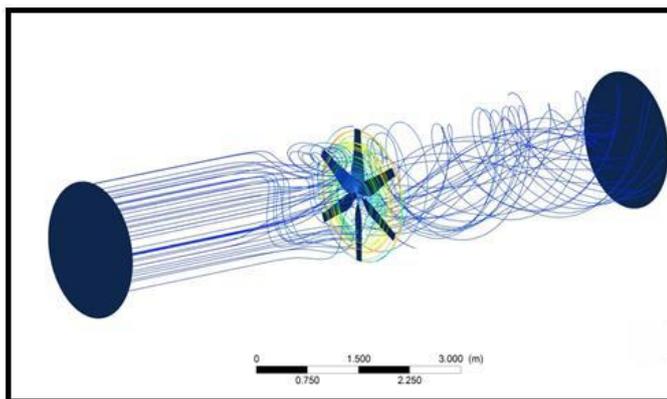


Figure 9.31 GOE 682 Streamlines

The 682 airfoil streamlines are much different from the other airfoils. From the above figure, we can see that the effect of the wind turbine blade is near the vicinity of the blade which indicates that the inlet velocity of the blade is low. The inlet velocity obtained here is 2.8 m/s.

9.5.2. CONTOURS OF VELOCITY MAGNITUDE

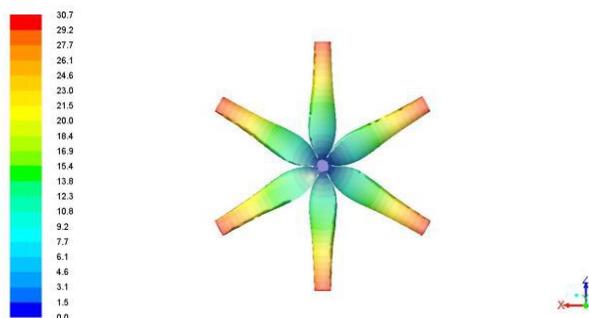


Figure 9.32 Velocity Contours- GOE 682

9.5.3. PRESSURE CONTOURS

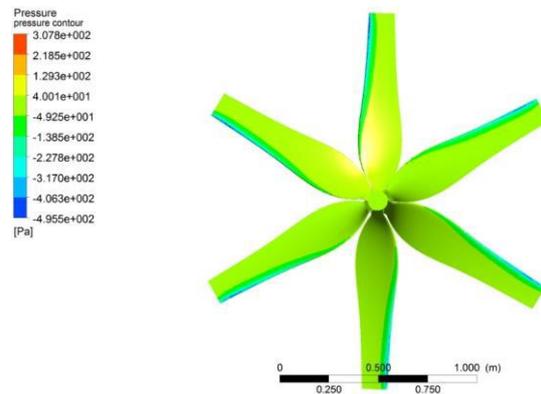


Figure 9.33 Pressure contours- GOE 682

9.6. COMPUTATIONAL RESULTS OF WIND TURBINE

An overview of the computational results adopted in this investigation is presented in this chapter. The CFD investigations resulted in some interesting observations. The figure indicates the average velocity of the wind hitting the turbine blade at the rated speed of the turbine.

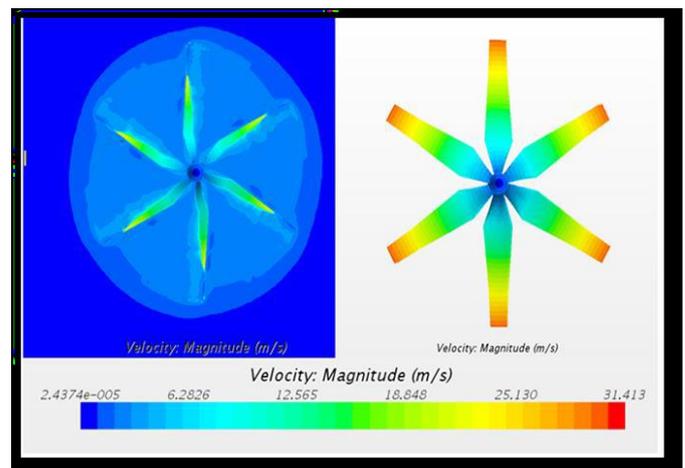


Figure 9.33 Distribution of Contour of velocity over the blade

As the result the velocity magnitude has a maximum value at the tip of the blades and is practically zero at the centre of the hub. The very fact is that all the blade tips show maximum velocity and this ensures that the turbine's axis passes through the center of the hub and is aligned properly without disturbing the orientation of the turbine. This investigation also concludes that the turbine is rotating in the anti-clockwise direction. Analysis on contour of pressure reveals that the pressure is exerted on the wind turbine blades by the incoming air.

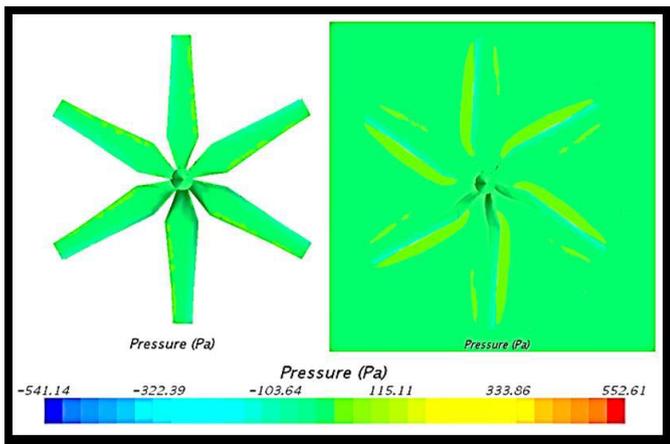


Figure 9.34 Distribution of contours of pressure over the blade

As seen in the pressure contour on the blade surface is in the range of 100 Pa and the visualization shows that the darker region is where the pressure is high and that it is rotating in the anti-clockwise direction. These pressure plots do not affect the computational power output adversely because they occur at the inner blade span, which contributes little to the torque in the blades. The velocity magnitude figure reveals the path of the wind when it hits the blades and passes through the turbine at a steady state. The path of the wind would vary at every second for a transient state analysis.

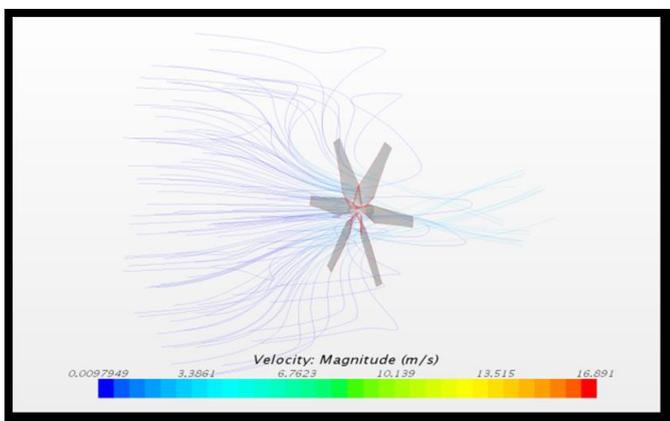


Figure 9.35 Velocity Streamline of the rotating blade

The streamlines tell us about the path of the wind as it hits the blade. This path varies every second. The streamlines of air pass through the wind turbine. We can see that the wind flow is concentrated towards the hub. As the turbine is rotating, the air in front of it is sucked in and pushed out. This will cause a pressure drop in front of the turbine. The blade angle is set in such a way that the starting torque is high. The streamline of air passes through the turbine blades as shown in Figure. The numerical solution, for the velocity at rated power is close to the experimental values. The high blade angle at the root provides high starting torque for the low wind speeds. From the streamline investigation, it is

clear that the wind speed averages around 3.4 m/s. After the solution is converged, the CFD post processing is run to obtain the various results.

10. CONCLUSION

Under typical design conditions, the rated power of 1kW was obtained by vigilant experimentation and was adequately validated with the investigations on CFD. The test results indicate that the blades were sufficiently efficient and it complied with the design considerations of the turbine. The wind tunnel tests were useful to select the blade angles by indicating the ratio of lift to drag coefficients at different sections along the length of the blade. The blade design and the performance of these blades in low wind speed conditions were observed at 300 rpm and simulation results ensure optimum power output at 3.4 m/s. It was witnessed and demonstrated that the blades were designed efficiently and the rated power output was obtained at a very low wind speed of 4.4m/s.

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